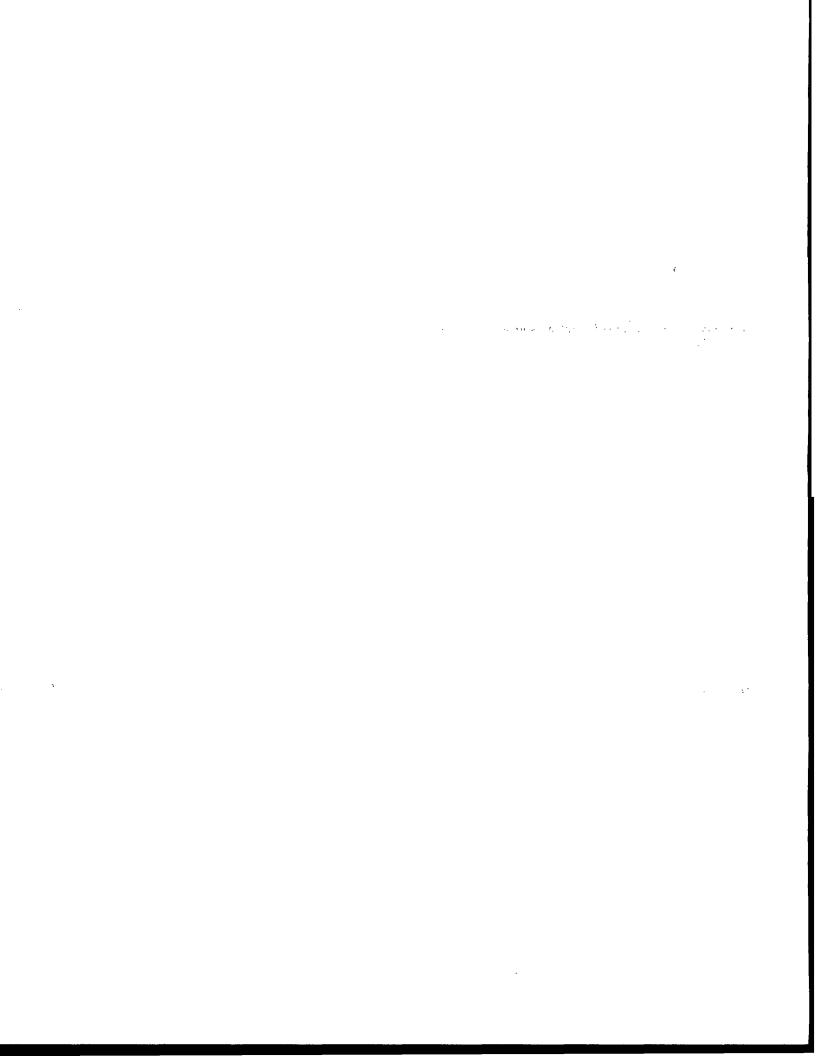
US Department
of Transportation
Federal Aviation
Administration

### MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Exp: 5/31/2018		
	For FAA Use Only	

	leral Aviation ninistration		•		-	-			-					ĺ
ins	structions	TONS: Print and dispos	ition of this	form.	es. See Title 14 This report is req	CF	R §	43.9, Part 43 by law (49 U.S	Appendix B, a S.C. §44701). I	and AC 4 ailure to	43.9- repo	1 (or sub ort can res	sequent revision there sult in a civil penalty fo	of) for r each
		Nationalit United	y and Regi		n Mark 60X				Serial No.	B-17	— 83			
1. Ai	rcraft	Make			Aircraft	С	or	npany	Model Kin	ıg Ai	r	S	B200	
2. O	wner		shown on EN MFO	-	ration certificate)				Address (As Address 4025 E City Colum Zip 68601	23RD ST	n reg	istration c	State NE	
							3. F	or FAA Use	Only					
	4. Ty	pe					5. L	Jnit Identifica	tion					
R	ераіг	Alteration	Uni	t		Mak	ke			Model			Serial No.	
		X	AIRFRAM	IE	Raytheon Airc	craft	Com	pany	(As describ	ed in Iten	n 1 a	bove)	BB-1783	
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Addres	6562 Cou	ntry Links Place			Nebrooke	- [		Foreign Certi	ficated Mechanic			C. Certific	cate No.	
City Zip	68601		untry United	States	State Nebraska	-	1		epair Station laintenance Orga	nization	$\dashv$	Αδ	&P 352898	8
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Exte per 1 App.	nded ran 14 CFR F B	ige fuel Part 43		Signa	ature/Date of Author	oriz	ed I	ndividual	Smell			2-28	2-2016	
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BY	FA	\A Designee		Repa	air Station	X	Ins	spection Auth	orization	Other (	Spec	eify)		
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#### NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets.	Identify with aircraft nation	onality and reg		e work completed.) King Air B200
		Nationality a	N360X nd Registration Mark	12-22-2016 Date
Aircraft Total Time 3009.3				
1) Installed Soft Touch Tires an altern 028-335-3 in accordance with Comm 03/07/2008.	nate set of main l outer Air Technolo	landing g ogy Top I	ear tires, Miche Drawing LE100	elin part number 00, Revision C Dated
2) Approval for this installation is from 14700 N. Airport Drive Scottsdale, Az	Z 85260.			r Technology, Inc.
<ol> <li>Updated Aircraft Weight and Balar</li> <li>Installed FAA Approved Airplane F Handbook.</li> </ol>	rce by adding 5 i Flight Manual Sup	os/tire at oplement	No.LE-10001 i	nto Pilot's Operating
				3
				ľ
	Additional Sheets	Are Attache	<del>i</del>	

King Air B200 S/N:BB-1783 Non-Transferable

May 05,2016

United States Of America

Bepartment of Transportation - Federal Abiation Administration

### Supplemental Type Certificate

Number SA00184LA

This Certificate issued to

Commuter Air Technology, Inc. 14700 N. Airport Drive Scottsdale, AZ 85260

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Psegulations. [For Certification Basis see Type Certificate Data Sheet A24CE]

Original Product Typo Certificate Number: A24CE

Mbake: Beech Aircraft Corporation

Model: 200, 200C, 200CT, 200T, A200, A200C, A200CT, B200, B200C, B200CT, B200T, B200GT, B200CGT

Description of Type Design Change Installation of Alternate Main Landing Gear Tires, Michelin Part Number 026-335-0 in accordance with Commuter Air Technology Top Drawing LE10000, Revision NC dated May 9, 1996, or later FAA approved revision. Commuter Air Technology, Inc. Pilot's Operating Handbook and FAA Approved Airplane Flight Manual Supplement Document No. LE-10001, FAA approved on May 14, 1996, or later FAA approved revision, is required as part of this installation.

Limitations and Conditions. This installation should not be incorporated in any aircraft unless it is determined that the interrelationship between this installation and any previously approved configuration will not introduce any adverse effect upon the airworthiness of the aircraft. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: 11/23/95

Date of issuance: May 14, 1996

Duto reissued:

Date amended . April 18, 2008

By direction of the Administrator

Manager, Cabin Safety/Mechanical and Environmental Systems Branch

Los Angeles Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

		,



US Department of Transportation Federal Aviation Administration

### MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only Office Identification

GL05

INSTRUCTIONS: Print or type all entries. See FAR 43.9 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

1. Aircraft	Beecher	aft				Model B200										
	Serial No.			<del></del>			Nationality and	Registratio	п Маг	rk						
	BB-1783			_			N360X	Biorram	, 1, 1, 1 u i	*** ·.						
	Name (As sho	wn on	registration certi	ificate)		vi.	Address (As sho	own on regi	stratio	n certificate)						
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1						*. •										
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							requirements and is	approved only	ly for the above described aircraft							
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VANDALIA, OI	Н 45377			H		facturer	1011		AAC							
D. I certify that t	he repair and/or alt	cration	made to the unit(s)	identified i	n Item	above and descr	ibed on the reverse	or attachme	nte her	eto have been m	ade in					
Accordance w my knowledge	nth the requirement	s of Pa	art 43 of the U.S. Fee	icral Aviat	ion Reg	ulations and that	the information fu	mished herei	n is tru	e and correct to t	he best of					
Date	<u>, , , , , , , , , , , , , , , , , , , </u>		<del></del>	Sign	nafure (	of Authorized In	dividual									
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Aviation Adminis	ithority given perso tration and is:	ns spe	cified below, the uni	it ideptified	in item	4 was inspected REJECT	in the manner pres	cribed by the	e Admii	nistrator of the F	ederal					
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#### NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircrast record. An alteration must be Compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

BEECHCRAFT B200

S/N: BB-1783

Installed a Bendix/King KGP-860 EGPWS.

#### Installed Items

1. Mid-Continent Terrain Awareness Control Unit (TACU) (p/n: MD41-1308) at Station 106.5.

2. Bendix/King KGP-560 Processor (p/n: 965-1198-005; SW 005) at Station 56.0. Referenced KGP-560 System Installation Manual P/N: 006-10611-0003, Rev. 3, dated June 16 2003, and STC SA00886WI-D.

3. Bendix/King Configuration Module (p/n: 700-1710-021) at Station 56.0.

→ This installation complies with TSO C151a Class B TAWS.

→ The KGP-560/860 is displayed on the existing Collins Radar/MFD.

→ The KGP-560/860 receives GPS position information from the existing Universal UNS-1K FMS, altitude information from a Collins ADC 85 air data computer, and provides audio alerts through the existing audio system.

> See page 2 for the Instructions for Continued Airworthiness.

→ Mounted the KGP 560/860 processor to the existing shelf at station 56.0. Fabricated a 11.75" x 4.5" plate of 2024-T3 Alclad aluminum and fastened it to existing shelf structure using (4) MS35206-246 screws, (4) AN960-8 washers, and (4) MS21044N08 firebrick nuts. The KGP-560 is attached to the plate using (4) MS21059L06 nutplates, (4) MS35206-230 screws, and (4) AN960-6 washers. The KGP 560 processor (combined weight of 1.5lbs.) is within the weight limits of the existing shelf.

An electrical load analysis was performed and found to be within 80% of the rated generator outputs. All equipment was installed using the Bendix/King KGP 560/860 System Installation Manual; AC25-10; AC25-16; AC25-23. Altered the equipment list and computed a new weight and balance.

A flight manual supplement dated FEB 2 4 2005, and the Bendix/King KGP-560/860 Pilot's Guide (P/N: 006-18254-0001, Rev 4, Jan 2004) are required for this approval and have been placed in the aircraft.

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)  BEECHCRAFT B200  S/N: BB-1783  N360X
DEECHCRAFT D200 ,, SITT. DD-1703 (1500A
KGP-560 Instructions for Continued Airworthiness:
1. See page 1.
2. See page 1.
<ol> <li>Refer to the Honeywell KGP 560/860 Pilot's Guide for operating procedures.</li> <li>N/A.</li> </ol>
5. If the MD41-1308 Annunciator/Switch Unit malfunctions or an annunciator bulb needs replaced, remove the unit and have it repaired by the factory. During each phase inspection of the aircraft, perform a Level 2 Self Test of the KGP-560/860 as defined in the Post Installation/Ground Test Checkout in Section 5.6 of the KGP-560 Installation Manual.
6. Refer to the Bendix/King KGP-560/860 Pilot's Guide for message annunciation descriptions. If the unit should require maintenance, removed the unit and have it repaired by an appropriately rated Honeywell approved Instrument Service Center.
7. Removal & replacement of the Bendix/King KGP-560/860 requires gaining access to the nose avionics bay and using a #2 Phillips Screwdriver to remove the mounting screws. After replacement perform a Level 2 Self Test of the KGP-560/860 as defined in the Post Installation/Ground Test Checkout in Section 5.6 of the KGP-560 Installation Manual
8. N/A.
9. N/A.
10. N/A.
11. N/A.
12. N/A.
13. N/A.
14. No additional overhaul time limitations.
15. N/A.
16. To revise these Instructions for Continued Airworthiness (ICA), submit a new FAA Form 337 along with the revised ICA to the local Flight Standard District Office for approval.
END
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Page 2 of 2

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#### US Department of Transportation Federal Aviation Administration

### MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only
Office Identificato E-FSDO-07
Wichita, Kansas

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

for	each s	such violation (Se	ection	901 Federal Aviat	ion Act	of 19	58).						j			
		Make F	TYAS	HEON AIRCRA	T CC	MPA	NY	Model	-	BEECH	в200					
1. Ai	rcraft	Serial No.		BB-178	3			Nationalii USA	ty and	Registratio N625						
		Name (As sh	own o	on registration cer	tificate	)		Address (	As sho	wn on regi	stration co	ertificate)				
		_						P.O. Box 85								
2. Ov	wner	F	RAYT.	HEON AIRCRAI	FT CC	)MPA	NY	Wich	ita,	Kansas	672	01-0085				
						3. F	or FAA Use O	nly								
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				Date		Approv	ing Inspector		fichita, KS							
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BY		AA Designee	х	Repair Station		Can	on Approved bada Airworthin	ess Group								
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	April 17, 2002 CRS BWVR672C CRS BWVR672C Dept. Insp.															

#### NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

#### 8. Description of Work Accomplished

(If more space is required, attach additional sheets, Identify with aircraft nationality and registration mark and date work completed.)

Installed a BFGoodrich Aerospace, BFGoodrich Avionics Systems, Inc. (BFG), integrated WX-1000+StormScope, Series II/SKY497 SkyWatch system complete this date, with reference to BFG StormScope, Series II Weather Mapping Systems WX-1000 Installation Manual No. 78-8051-9150-5 (Rev. D, 13 November 2001), BFG SkyWatch Traffic Advisory System, SKY497 Installation Manual No. 009-10800-001 (Rev. C, 23 Feb. 2001), Raytheon Aircraft Company Avionics Drawings for aircraft S/N BB-1783, AC 43.13-1B Ch. 6, Secs. 3 and 4; Ch. 7, Secs. 1 thru 5; Ch. 11, Secs. 1,3,8 thru 13, 15 and 17; Ch. 12, Secs. 1 and 2, and AC 43.13-2A, Chapters 1,2,3,11 and 13. (The installation of the typical SkyWatch components used in this application has been previously FAA-approved per S.T.C. SA00733CH, as installed in a Beech Model 58P Baron.) This Alteration complies with the safety requirements of FAR 23.1301 and FAR 23.1431; copies of installation data and wiring prints are on file at this Station.

The integrated system consists of (one of each) WX-1000/SKY497 StormScope/SkyWatch Control Display Unit (P/N 78-8060-5900-8/TSO C110a, C113), located in the lower/RH position of the pilot's flight instrument panel at station 101" aft datum, the StormScope WX-1000+ Processor Unit (P/N 78-8051-9160-4/TSO C110a; SV 3.27) and mount, located on the existing upper equipment shelf in the aft fuselage compartment, with the SkyWatch TRC497 Transmitter Receiver Computer (P/N 805-10800-001/TSO C147; SV 1.6) and mount, at stations 372" and 378" aft datum, respectively, the StormScope Series II Antenna (P/N 78-8051-9200-8/TSO C110a) installed with a typical/standard-equipment (King Air) doubler on the lower/RH surface of the aft fuselage at station 383" aft datum and the SkyWatch NY164 Directional Antenna (P/N 805-10890-001/TSO C118) installed on the upper surface of the forward fuselage at station 182" aft datum, using a doubler configuration that has been structurally-approved per FAA Form 8110-3. The system is provided analog 'heading' information from the 'Compass-1' system synchro repeater output (Bootstrap) of the copilot's Collins RMI-30 and 'altitude' information, in ARINC 429-format, from an auxiliary data output of the existing Collins ADC-85 Air Data Computer. The SkyWatch audio output has been connected to an auxiliary/unmuted input of the existing pilot's and copilot's DB Systems Model 438 Audio Amplifiers. A switch/annunciator, with the legends 'SKYWATCH' (or) 'STORMSCOPE' has been added to the pilot's flight instrument panel, that provides selection and indication of the information being presented in the Control Display Unit. The 28VDC primary power for each system is supplied from individual Circuit Breakers located in, and powered by, the 'Avionics'-section of the flight compartment RH sidewall (Main) DC Power Distribution Panel. The Circuit Breakers are identified as 'STORMSCOPE' and 'SKYWATCH'; the electrical ratings are 2 Amp. and 5 Amp., respectively. 26VAC, 400Hz Instrument Reference power is supplied to the RMI-2 (Compass-1) 'Bootstrap' transmitter, the SS Processor, and SW Computer, from a 1 Amp. (MDL-1/'Slow-Blow') fuse located on, and supplied power by, Avionics Junction Box-1 (in the forward/nose equipment compartment, RH side), identified as 'SS/SW REF'. (A load analysis has been performed to verify the system power sources are suitable and adequate for the new equipment.) A panel has been provided in the aft fuselage that contains two (GMW-1/Fast-Acting 1 Amp.) fuses, identified as 'STORMSCOPE' and 'SKYWATCH', that supply 26VAC Reference power to the remote units. The panel also contains a switch to enable 'SKYWATCH'-system operation with the StormScope Processor removed and a wiring 'jumper' receptacle to enable 'STORMSCOPE'-system operation with the 'SkyWatch' Computer removed.

The systems' operations have been ground-checked per applicable portions of the BFG Installation Manuals and the Airplane Flight Manual Equipment List and Weight/Balance data have been revised to include the new equipment. A StormScope, Series II Pilot's Guide, P/N 75-0299-7690-1 (Rev. C, 4/6/00) and a SKYWATCH Traffic Advisory System Pilot's Guide, P/N 009-10801-001 (Rev. B, 6/6/00) have been provided for the aircraft flight crew.

INSTRUCTIONS for CONTINUED AIRWORTHINESS: No scheduled maintenance is required to ensure system Continued Airworthiness, however, it is recommended that the StormScope system operation be checked out and verified satisfactory at least annually, especially prior to each thunderstorm season. At regular inspection intervals, perform the periodic maintenance procedures for all system components as detailed in the SKY497 Installation Manual Chapter 4: Maintenance, Paragraph 4.3.1 thru 4.3.3 (copy attached).

(	1	7	APR	200	1	)	The	FAA-approved	Airplane	Flight	Manual	Sup	plement	for	the	`SkyWat	ch' sys	tem ha	s
bee	en a	adde	ed to	the	A	irpl	ane	Flight Manual	'Suppleme	ents'-se	ection:	The	integra	ated	Stor	mScope/	SkyWatcl	ı syste	m
is	sat	is:	facto	ry :	Eor	fli	ght	operations as	detailed	in the	Supple	nent.	•						

END

Additional	Sheets Ar	e Attached
Additional	SHEETS AL	e Attacheu

## UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION

# STANDARD AIRWORTHINESS CERTIFICATE

2. MANUFACTURER AND MODE 4. CATEGORY 3. AIRCRAFT SERIAL 1. NATIONALITY AND NUMBER REGISTRATION MARKS RAYTHEOM/AIRCRAFT COMPAN NORMAL **BBA1**783 N360X

5. AUTHORITY AND BASIS FOR ISSUANDE

This airworthiness certificate is issued pursuant to the reference aircraft to which issued has been inspected and representation of the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civit Aviation except as noted herein.

Exceptions:

6. TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a fermination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43 and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United

States.

DATE OF ISSUANCE

DESIGNATION NUMBER

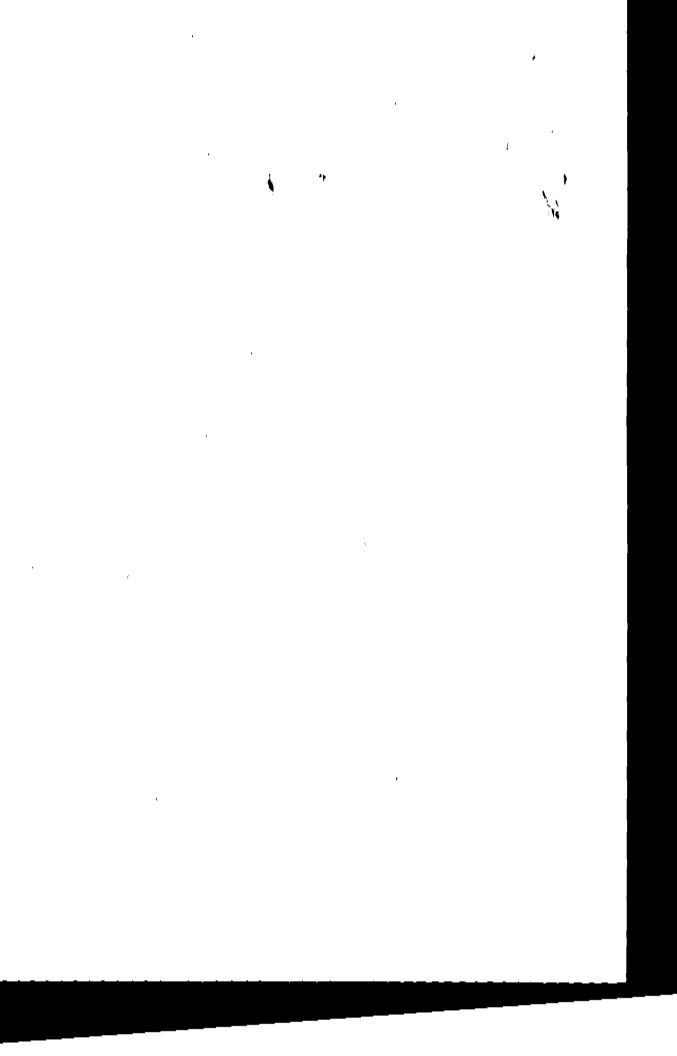
R 10/9/2001

GL07 CMH FSDO

Any alteration, reproduction, or misuse of this certificate may be purishable by a fine not exceeding \$1,000, or imprisonment not exceeding \$ years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

FAA Form 8100-2 (8-82)

U.S. GPO-2001 - 668-455



# UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION ATA NO A DO A DOMAGO STOURING OF THE STATE OF T

### STANDARD AIRWORTHINESS CERTIFICATE

1. NATIONALITY AND REGISTRATION MARKS.

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2. MANUFACTURER AND MODEL

3. AIRCRAFT SERIAL NUMBER

4. CATEGORY

N625TA

RAYTHEON AIRCRAFT COMPANY B200

BB-1783

NORMAL

#### 5. AUTHORITY AND BASIS FOR ISSUANCE

This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.

Exceptions:

NONE

\$

#### 6. TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE

FAA REPRESENTATIVE

DESIGNATION NUMBER

10-09-2001

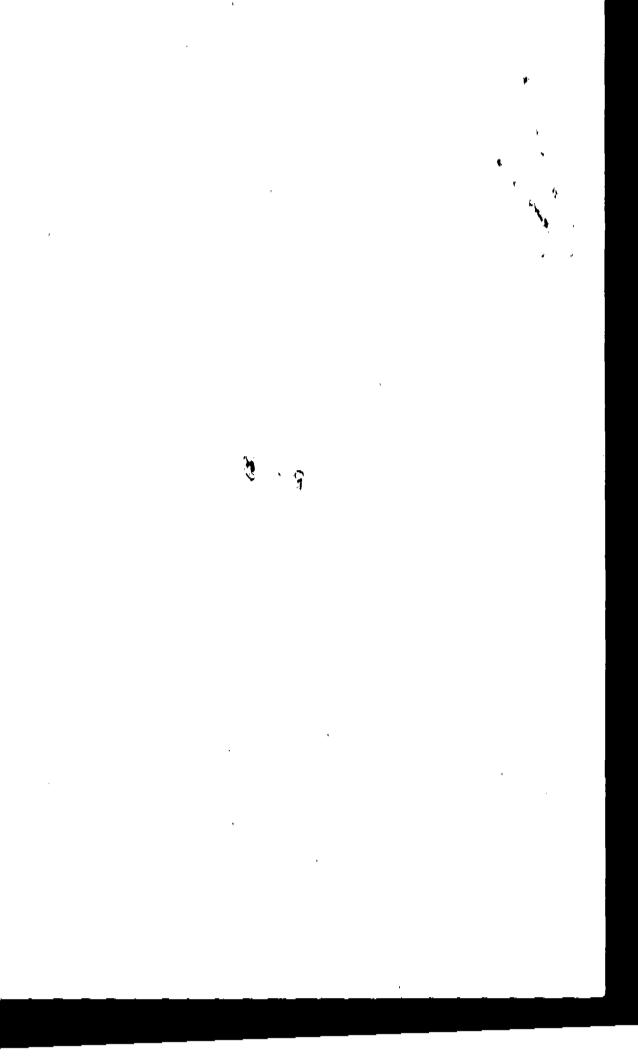
Robert Alliband

DOA-230339-CE

Any alteration, reproduction, or misuse of this certificate may be punishable by a fine not exceeding \$1,000, or imprisonment not exceeding 3 years, or both. THIS CERTIFICATE MUST BE DISPLAYED IN THE AIRCRAFT IN ACCORDANCE WITH APPLICABLE FEDERAL AVIATION REGULATIONS.

FAA Form 8100-2 (8-82)

# U.S. G.P.O.:1999 769-017



U.S. Departmont of Transportation Aviation Administration	ion Ion	CEPTIEICATE												INSTRUCTIONS - Print or type. Do not write in shaded areas. These are for use only. Submit original only to an authorized FAA Representative. If additi space is required use an attachment. For special flight permits complete Sec II, VI and VII as applicable.								
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I, AIRCRAFT DESCRIPTION	5. AI	RCR/		ERIA	T NO	<b>)</b> .	6. ENGI	NE B	UILDERS NAM	E (Ma	ke)		7. ENGINE MODEL DESIGNATION PT6A-42					•	52032			
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	RAYTHEON AIRCRAFT COMPANY													P.O.	Во	x 85 W	ichi	ta, KS	67201			
	B. AIRCRAFT CERTIFICATION BASIS (Check applicable blocks and complete items as indicated,																					
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IV. INSPECTION AGENCY VERIFICATION	5	ŀ	AIRC	RAFT	1AM	NUFACTURER (Give na	me of firm)	)														
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	10-09-2001 CE-43							Robert Alliband; DC						DOA+230339-CE /								

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and the second second		(beniupen nerw berbelliA)		ŧ	B. Current Operating Limitations Attache		VIII. AIRWORTHINESS DOCUMENTATION (FAA/DESIGGNEE uso only)			
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G. Statement of conformity, FAA Form 8130-9 (Attach when required)			<u> </u>	9.19 notizes SECTION 91.9	co on a Spirithing Limitations and Markings in co					
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IDDRESS				A. DESCRIPTION OF AIRCRAFT REGISTERED OWNER						
SIGNANDIS				DATE OF APPLICATION MAME AND TITLE (Print or lype)						
		<del>-</del>		C. GIVE QUANTITY OF CERTIFICATES REQUIRED FOR OPERATING NEEDS						
				TION SYSTEM	TYPE CERTIFICATE ONLY APPROVED PRODUCTION INSPEC		VI. PRODUCTION FLIGHT TESTING			
PRODUCT CERTIFICATE (Give production certificate number)							OIL			
B. PRODUCTION BASIS (Check applicable items)										
RDSESS MORESS				×		MAN				

A. MANUFACTURER

# UNITED STATES OF AMERICA DEPARTMENT OF TRANSPORTATION—FEDERAL AVIATION ADMINISTRATION

### STANDARD AIRWORTHINESS CERTIFICATE

NATIONALITY AND REGISTRATION MARKS
RAYTHFON ATRICRAFT COMPANY
N625TA

2. MANUFACTURER AND MODEL 3. AIRCRAFT SERIAL NUMBER

4. CATEGORY
NUMBER

RP-1783

5 AUTHORITY AND BASIS FOR ISSUANCE

This airworthiness certificate is issued pursuant to the Federal Aviation Act of 1958 and certifies that, as of the date of issuance, the aircraft to which issued has been inspected and found to conform to the type certificate therefor, to be in condition for safe operation, and has been shown to meet the requirements of the applicable comprehensive and detailed airworthiness code as provided by Annex 8 to the Convention on International Civil Aviation, except as noted herein.

Exceptions:

MOVE

#### 6 TERMS AND CONDITIONS

Unless sooner surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator, this airworthiness certificate is effective as long as the maintenance, preventative maintenance, and alterations are performed in accordance with Parts 21, 43, and 91 of the Federal Aviation Regulations, as appropriate, and the aircraft is registered in the United States.

DATE OF ISSUANCE

FAA REPRESENTATIVE

DESIGNATION NUMBER

10-09-2001

Robert Alliband

DOA-230339-CE

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